





## CHALLENGE FOR JUPITER SYSTEM EXPLORATION

HANNA ROTHKAEHL

12 09 2023 PTA TORUŃ

JUICE JUpiter ICy Moon Explorer ESA L class mission

## 14 04 2023 14:14 CET







The JUICE mission will address two themes of ESA's Cosmic Vision programme: 1.What are the conditions for planet formation and emergence of life? 2. How does the Solar System work?

The JUpiter ICy moons Explorer (JUICE) will perform detailed investigations of Jupiter and its system in all their inter-relations and complexity with particular emphasis on Ganymede as a planetary body and **potential habitat**.

Investigations of Europa and Callisto would complete a comparative picture of the Galilean moons.



Dimensions (stowed for launch): 4.09 x 2.86 x 4.35 m Dimensions (deployed in orbit): 16.8 x 27.1 x 13.7 m Dry mass (without fuel): 2420 kg. This includes the 'payload adapter' that connects the satellite to the launcher. Amount of propellant (full tank): 3650 kg.

## Instrument payload mass: 280 kg



**Solar panels**: Juice has a distinctively shaped solar array – two 'wings' of panels in a cross-like formation. Overall, these wings are made up of ten 2.5 x 3.5 m panels (five on each side) with a total area of 85 m<sup>2</sup>









## You tube Astronanrium 160















# JUPITER ICY Moon Explorer ESA L class mission 2023

5 04 -25 04 2023 lauched window

JANUS - Camera system MAJIS - Moons and Jupiter Imaging Spectrometer UVS - UV imaging Spectrograph SWI - Sub-millimeter Wave Instrument GALA - GAnymede Laser Altimeter RIME - Radar for Icy Moons Exploration J-MAG - A magnetometer instrument for JUICE RPWI - Radio & Plasma Wave Investigation 3GM - Gravity & Geophysics of Jupiter and Galilean Moons PRIDE - Planetary Radio Interferometer & Doppler Experiment





## First photo

## **RPWI** - Radio & Plasma Wave Investigation



**19-23 April** RPWI Sensors release and first signals detected, FFT – Full Functional Tests



Jan-Erik Wahlund (PI) Swedish Institute of Space Physics

Baptiste Cecconi (Co-PI) LESIA-Observatoire de Paris, 5 place Jules

Yasumasa Kasaba (Co-PI) Department of Geophysics, Tohoku University

Hanna Rothkaehl (Co-PI) Space Research Centre PAS

Ondrej Santolik (Co-PI) Institute of Atmospheric Physics

Ingo Müller-Wodarg (Co-PI) Space and Atmospheric Physics group Imperial College London, London, UK RPWI makes use of several different sensors and receivers. Altogether, the instrument uses sensors and 3 receivers, which cover a wide frequency range, from DC up to 45 MHz. There are 4 Langmuir probes (LP-PWI) for plasma and electric field measurements, a search coil magnetometer (SCM) with 3 coils for magnetic fields measurement, and 3 radio antennas (RWI). Thus, the RPWI sensors provide complete measurements of the electric and magnetic field vectors.







Electric field vector, $\delta E(f)$ DC - 1.5 MHz Dust detection. Polarization with $\delta B(f)$ . Polarization with $\delta B(f)$ . Polarization with $\delta B(f)$ . Polarization with $\delta B(f)$ . Polarization with $\delta B(f)$ . Spectral (differential): $<1 \mu V m / (Matrument)$ (Calibration for SC influence needed in flight)Electric field vector, $\delta E(f)$ $80  kHz - 45  MHz$ Wave vector ( $\delta k$ ) Polarization $10  nV/m / Hz$ (at 10 MHz) $-1^{\circ}$ (at 5 MHz) (Direction finding calibration needed in- flight)Magnetic field vector, $\delta B(f)$ $0.1Hz - 20  kHz$ Polarization $8  pT / Hz$ (at 1 Hz) $0.6  pT / Hz$ (at 10 Hz) $0.6  pT / Hz$ (stout) $0.6  pT / Hz$ (at 10 Hz) $0.6  pT / Hz$ (at 10 Hz) $0.6  pT / Hz$ (stout) $10.6  pT / Hz$ (stout) $0.6  pT / Hz$ Electron density (Ne) $10^4 - 10^5  cm^3$ $1 - 20  eV$ Ion drift speed (Vdi) $0.1 - 20  eV$ Ion drift speed (Vdi) $0.1 -$	PWI Measured Quantity	Range/Product	Error/Sensitivity
Electric field vector, $\delta E(f)$ 80 kHz – 45 MHz Wave vector ( $\delta k$ ) Polarization10 nV/m/\ <sup>1</sup> Hz (at 10 MHz) $-1^{\circ}$ (at 5 MHz) (Direction finding calibration needed in- flight)Magnetic field vector, $\delta B(f)$ 0.1Hz – 20 kHz Polarization with $\delta E(f)$ . Poynting flux with $\delta E(f)$ . 20 fT/\ <sup>1</sup> Hz (at 10 Hz) 0.06 pT/\ <sup>1</sup> Hz (s00Hz-10kHz)Electron density (Ne) $10^{4} - 10^{5}$ cm <sup>-3</sup> <10% (>100em <sup>-3</sup> ) Leak current $\approx 0.5$ pA <100Hz	Electric field vector, δ <b>E</b> (f)	DC - 1.5 MHz Dust detection. Polarization with $\delta B(f)$ . Poynting flux with $\delta B(f)$ .	$\begin{array}{l} DC:\approx 0.9 \ \mu V/m \ (instrument) \\ (Calibration \ for \ S/C \ influence \ needed \ in \\ flight) \\ Spectral \ (differential): \\ <1 \mu V/m/\sqrt{Hz} \ (>500 Hz) \end{array}$
Magnetic field vector, $\delta B(f)$ 0.1Hz - 20 kHz Polarization with $\delta E(f)$ . Poynting flux with $\delta E(f)$ . Poynting flux with $\delta E(f)$ . Poynting flux with $\delta E(f)$ .8 pT/√Hz 0.6 pT/√Hz (at 10 Hz) 20 fT/√Hz (at 100 Hz) 20 fT/√Hz 	Electric field vector, δE(f)	80 kHz – 45 MHz Wave vector (δk) Polarization	10 nV/m/√Hz (at 10 MHz) ~1° (at 5 MHz) ~10% (at 5 MHz) (Direction finding calibration needed in- flight)
Electron density (Ne) $10^4 - 10^5 \text{ cm}^{-3}$ $<10\% (>10 \text{ cm}^{-3})$ Leak current $\approx 0.5 \text{pA} < 100 \text{Hz}$ Density fluctuations ( $\delta n/n$ )DC - 10kHz $<5\%$ below 1kHz $\delta E$ or $\delta n$ interferometry $<1000 \text{ km/s}$ Phase accuracy $<1^\circ$ for $<20 \text{ kHz}$ . Phase response correctable.Ion density (Ni) $1-10^5 \text{ cm}^{-3}$ $<20\%$ Electron temperature (Te) $0.01 - 20 \text{ eV}$ $<20\%$ Ion drift speed (Vdi) $0.1-200 \text{ km/s}$ $<20\%$ Ion temperature (Ti) upper constraints $0.02 - 20 \text{ eV}$ Constrained by $Spacecraft potential (Usc)\pm 80 \text{ V}<10\%Integrated solar EUV flux<1 \text{ Hz}Res. 0.003 \text{ Gphotons/cm}^2/\text{s}PSSR: Dynamic range incl. processing gain85 \text{ dB}<10\%PSSR: Max Ice depth / Resolution<20 \text{ km} (depend on ice conductivity/salinity)<1 \text{ km}$	Magnetic field vector, $\delta \mathbf{B}(\mathbf{f})$	0.1Hz – 20 kHz Polarization with δE(f). Poynting flux with δE(f).	8 pT/√Hz (at 1 Hz) 0.6 pT/√Hz (at 10 Hz) 0.06 pT/√Hz (at 100 Hz) 20 fT/√Hz (500Hz-10kHz)
Density fluctuations ( $\delta n/n$ )DC - 10kHz<5% below 1kHz $\delta E$ or $\delta n$ interferometry<1000 km/s	Electron density (Ne)	$10^{-4} - 10^5 \text{ cm}^{-3}$	<10% (>10cm <sup>-3</sup> ) Leak current ≈0.5pA <100Hz
$\delta E$ or $\delta n$ interferometry $20^{\circ}$ Forming $20^{\circ}$ Forming $20^{\circ}$ Forming $\delta E$ or $\delta n$ interferometry $<1000 \text{ km/s}$ Phase accuracy <1° for <20kHz. Phase response correctable.Ion density (Ni) $1-10^{5} \text{ cm}^{-3}$ $<20\%$ Electron temperature (Te) $0.01 - 20 \text{ eV}$ $<20\%$ Ion drift speed (Vdi) $0.1-200 \text{ km/s}$ $<20\%$ Ion temperature (Ti) upper constraints $0.02 - 20 \text{ eV}$ Constrained by $Spacecraft potential (Usc)\pm 80 \text{ V}<10\%Integrated solar EUV flux<1 \text{ Hz}Res. 0.003 \text{ Gphotons/cm}^2/sPSSR: Dynamic range incl. processing gain85 \text{ dB}<1 \text{ km}$	Density fluctuations (\deltan/n)	DC – 10kHz	<5% below 1kHz
Ion density (Ni) $1-10^5 \text{ cm}^{-3}$ $<20\%$ Electron temperature (Te) $0.01 - 20 \text{ eV}$ $<20\%$ Ion drift speed (Vdi) $0.1-200 \text{ km/s}$ $<20\%$ Ion temperature (Ti) upper constraints $0.02 - 20 \text{ eV}$ Constrained by $< mV_{di}^2/2e$ Spacecraft potential (Usc) $\pm 80 \text{ V}$ $<10\%$ Integrated solar EUV flux $<1 \text{ Hz}$ Res. $0.003 \text{ Gphotons/cm}^2/s$ PSSR: Dynamic range incl. processing gain $85 \text{ dB}$ $<10\%$ PSSR: Max Ice depth / Resolution $<20 \text{ km}$ (depend on ice conductivity/salinity) $<1 \text{ km}$	δE or δn interferometry	<1000 km/s	Phase accuracy <1° for <20kHz. Phase response correctable.
Electron temperature (Te)0.01 - 20 eV<20%Ion drift speed (Vdi)0.1-200 km/s<20%	Ion density (Ni)	$1-10^5 \mathrm{cm}^{-3}$	<20%
Ion drift speed (Vdi)       0.1–200 km/s       <20%	Electron temperature (Te)	0.01 – 20 eV	<20%
Ion temperature (Ti) upper constraints $0.02 - 20 \text{ eV}$ Constrained by $< mV_{di}^2/2e$ Spacecraft potential (Usc) $\pm 80 \text{ V}$ $<10\%$ Integrated solar EUV flux $<1 \text{ Hz}$ Res. $0.003 \text{ Gphotons/cm}^2/s$ PSSR: Dynamic range incl. processing gain $85 \text{ dB}$ $<20 \text{ km}$ (depend on ice conductivity/salinity)	Ion drift speed (Vdi)	0.1–200 km/s	<20%
Spacecraft potential (Usc)     ±80 V     <10%       Integrated solar EUV flux     <1 Hz	Ion temperature (Ti) upper constraints	0.02 - 20  eV	Constrained by $\langle mV_{di}^2/2e$
Integrated solar EUV flux       <1 Hz	Spacecraft potential (Usc)	±80 V	<10%
PSSR: Dynamic range incl. processing gain       85 dB         PSSR: Max Ice depth / Resolution       <20 km (depend on ice conductivity/salinity)	Integrated solar EUV flux	<1 Hz	Res. 0.003 Gphotons/cm <sup>2</sup> /s
PSSR: Max Ice depth / Resolution     <20 km (depend on ice conductivity/salinity)	PSSR: Dynamic range incl. processing gain	85 dB	
	PSSR: Max Ice depth / Resolution	<20 km (depend on ice conductivity/salinity)	<1km



Instrument RPWI (Instrument Radio & Plasma Wave Investigation)

Pierwsze pomiary dokonane w przestrzeni kosmicznej daleko poza magnetosferą Ziemi, pokazały że przyrząd działa dobrze !.



In prep, The Radio & Plasma Wave Investigation (RPWI) for the JUpiter ICy moons Explorer (JUICE) w Space Sci Rev 2023



# First evidence of a solar radio emission 13 July



Left: Raw data from RWI Right: Filtered data

- The time axis is about minutes. RWI takes 5 spectra every 63 seconds. Each set takes 8 seconds.
- 32 samples/bin (4kHz)
- 512 samples/bin (250Hz) would have been desirable





DETECTION OF GANYMEDE **MAGNETOSPHERE & IONOSPHERE** Galileo PWS Ganymede 1 Flyby June 27, Day 179, 1996 →Ganymede Magnetosphere **Jovian Radio Emissions** (CM<sup>-3</sup>) Electric **ECH Bands Radio Emissions** DENSITY Whistler Modes NUMBER n, ELECTRON qв

10<sup>6</sup>

10<sup>5</sup>

10<sup>4</sup>

10<sup>3</sup>

10<sup>2</sup>

 $10^{1}$ 

10<sup>4</sup> V

10<sup>3</sup>

10<sup>2</sup>

 $10^{1}$ 

EΤ

R

٥n<sub>III</sub>

Lat





 $N_{e} \approx 200-300 \text{ cm}^{-3}$  @ 260 km [Gurnett et al., 1996]

 $T_i \approx 1.3 \text{ eV}$ [Frank et al., 1997]





## ICY MOON CONDUCTIVITIES & ELECTRIC CURRENTS

- Determine the electric conductivity of the ionospheres
  - Assess their **role in supporting MHD-dynamo generated current systems** induced by the rotating and variable Jovian magnetosphere
  - Assess how these currents may couple inductively to sub-surface oceans
- Monitor electric acceleration structures at magnet flux tubes connected to Ganymede's auroral regions.
- $\sigma_{\rm H} \approx$   $\sigma_{\rm P} \sim en_e/(2B) \sim 10^{-4}$ -10<sup>-3</sup> mho near surface
- $j \ge \sigma \ E \sim 0.1 \ \mu A/m^2$
- $I \ge 100 \text{ kA}$  through ionosphere?
- Or through salty sub-surface ocean?







## IONIZATION, HEATING AND DYNAMICS OF EXOSPHERES/IONOSPHERES

### **RPWI WILL:**

- MONITOR PLASMA DENSITIES 10<sup>-4</sup>
   10<sup>5</sup> CM<sup>-3</sup> (MS RESOLUTION)
- LOCATE (ELECTRON) HEATING REGIONS IN THE DENSE PLASMA (>0.1 CM<sup>-3</sup>)
- DETERMINE **EXB** CONVECTION
   AND BULK ION DRIFT SPEED
- MONITOR THE SIZE AND MASS
  DISTRIBUTION OF A POSSIBLY
  EXISTING CHARGED DUST
  COMPONENT
- MONITOR DUST-PLASMA
   INTERACTIONS







The induction process coupling the local space plasma where the spacecraft is situated to the electric currents in a conductive sub-surface ocean. The red parameters are measured in orbit (or during flybys) of the icy moons by RPWI and J-MAG onboard JUICE. The quasi-periodic magnetospheric induction disturbance dominantly from Jupiter's magnetodisc propagates down to the sub-surface ocean in an Alfvénic manner, and there generates an induction current. The ocean (e.g., tidal) motion also contributes to an electric current. The total induction response in the ocean propagates outward through the ionosphere to the magnetosphere, where the JUICE spacecraft measures the disturbed electric and magnetic fields at many frequencies (days, hours). Various other electric current sources, such as ionospheric electric currents, also contribute to the measured electric and magnetic fields and must be separated out from the contribution of the sub-surface ocean. For instance, the induction response itself generates electric currents in the ionosphere, and for Callisto, this ionospheric induction response can dominate the measurements.



## A TRIO OF MISSIONS: JUNO, JUICE AND EUROPA CLIPPER

Their destination may be the same, but Juno, Juice and Europa Clipper are all unique missions with different goals and instruments. Juno's discoveries are being used to optimise plans for Juice and Europa Clipper.



## JUNO

Target: Jupiter

Arrival: 2016

**Special skill:** a polar orbit that goes very close to Jupiter, allowing deep mapping of its gravity and magnetic fields

## JUICE

Targets: Jupiter, Ganymede, Europa, Callisto

### Arrival: 2031

**Special skill:** observing Jupiter and its icy moons to provide a complete view of habitable conditions in the Jovian system

## **EUROPA CLIPPER**

Target: Europa

Arrival: 2030

**Special skill:** investigating the potential for life on Europa; helping select a landing site for a future Europa lander

### Start 10 2024

Jupiter 04 2030

Ultraviolet Spectrograph/Europa (Europa-UVS) Principal Investigator: Dr. Kurt Retherford, Southwest Research Institute, San Antonio

Detect the likely presence of water plumes erupting from Europa's surface, including small plumes, and to provide valuable data about the composition and dynamics of Europa's rarefied atmosphere.

### Europa Imaging System (EIS)

- Principal Investigator: Dr. Elizabeth Turtle, Johns Hopkins University Applied Physics Laboratory
- Wide and narrow angle cameras to map most of Europa at better than 100 m resolution, and to provide images of areas of Europa's surface at up to 100 times higher resolution.

### Mapping Imaging Spectrometer for Europa (MISE)

- Principal Investigator: Dr. Diana Blaney, Jet Propulsion Laboratory, California Institute of Technology
- Probe the composition of Europa, identifying and mapping the distributions of organics, salts, acid hydrates, water ice phases, and other materials to determine the habitability of Europa's ocean.

### Europa Thermal Emission Imaging System (E-THEMIS)

- Principal Investigator: Dr. Philip Christensen, Arizona State University, Tempe
- Provide high spatial resolution, multi-spectral thermal imaging of Europa to help detect active sites, such as potential vents erupting plumes of water into space.

### Radar for Europa Assessment and Sounding: Ocean to Near-surface (REASON)

- Principal Investigator: Dr. Donald Blankenship, University of Texas, Austin
- Characterize and sound Europa's icy crust from the near-surface to the ocean, revealing the hidden structure of Europa's ice shell and potential water within.











## Europa Clipper In Situ Investigations

### Europa Clipper Magnetometer (ECM) - Project provided instrument

- Team Leader: Dr. Margaret Kivelson, University of Michigan, Ann Arbor
- To measure the magnetic field near Europa and infer the location, thickness and salinity of Europa's subsurface ocean using multi-frequency electromagnetic sounding.

### Plasma Instrument for Magnetic Sounding (PIMS)

- Principal Investigator: Dr. Joseph Westlake, Johns Hopkins University Applied Physics Laboratory
- In conjunction with a magnetometer, is key to determining Europa's ice shell thickness, ocean depth, and salinity by correcting the magnetic induction signal for plasma currents around Europa.

### MAss SPectrometer for Planetary EXploration/Europa (MASPEX)

- Principal Investigator: Dr. James Burch, Southwest Research Institute, San Antonio
- To determine the composition of the surface and subsurface ocean by measuring Europa's extremely tenuous atmosphere and any surface material ejected into space.

### SUrface Dust Mass Analyzer (SUDA)

- Principal Investigator: Dr. Sascha Kempf, Univ. Colorado, Boulder
- To measure the composition of small, solid particles ejected from Europa, providing the opportunity to directly sample the surface and potential plumes on low-altitude flybys.

### Gravity and Radio Science (G/RS) – Project provided instrument

- Team Leader: Dr. Erwan Mazarico, GSFC, Greenbelt, MD
- Using the telecommunications system to perform gravity and radio science experiments to enhance the science return of the Europa Clipper mission.

















### EUROPEAN PARTNERS

Many agencies, organisations and companies have contributed to the development of Juice. This map highlights the main contributing ESA Member States and their funding agencies. Prime contractor for the building of Juice is Airbus.

### Austria Austrian Research Promotion Agency

Belgium Belgian Science Policy Office

**Czech Republic** Department of Research and Development, Ministry of Education

France National Centre for Space Studies (CNES)

Germany German Space Agency at DLR

Greece Academy of Athens

Hungary Centre for Energy Research **Italy** Italian Space Agency (ASI)

**Poland** Ministry of Entrepreneurship and Technology

**Spain** Ministry of Economy and Competitiveness

Sweden Swedish National Space Agency

Switzerland Swiss Space Office

United Kingdom UK Space Agency

### Beyond Europe:

United States National Aeronautics and Space Administration (NASA)

•eesa

Japan Japan Aerospace Exploration Agency (JAXA)

Israel Israel Space Agency (ISA)

4